Amendments to the Claims:

1-10 (canceled)

11. (currently amended) A turbine blade, comprising:

a blade leaf arranged along a blade axis having a blade tip, a root opposite the tip, a suction side and a pressure side;

a platform region arranged at the root of the blade leaf;

a platform arranged at the platform region having a width and ~~extends~~ extending transversely with respect to the blade axis;

a first platform wall arranged along the transition from the blade leaf to the platform that does not structurally support the blade leaf and has along the width of the platform an aerodynamic shape; and

a second platform wall arranged in the platform region that structurally supports and is continuous with the blade leaf and has, along the width of the platform, a set-back step with respect to the first platform.

12. (previously presented) The turbine blade as claimed in claim 11, wherein an interspace for cooling the platform is formed between the aerodynamic shape of the first platform wall and the set-back step of the second platform wall.

13. (currently amended) The turbine blade as claimed in claim 12, wherein the interspace has a uniform height defined along the entire width of the platform by a height of the set-back step.

14. (previously presented) The turbine blade as claimed in claim 11, wherein the second platform wall thickness is greater the first platform wall thickness.

15. (previously presented) The turbine blade as claimed in claim 11, wherein the second platform wall has a plurality of cooling passages per unit area greater along the transition from the blade leaf to the platform than in the remainder of the platform region.

16. (previously presented) The turbine blade as claimed in claim 11, wherein the first platform wall is formed by a resilient elastic sheet metal part arranged adjacent the blade leaf.

17. (previously presented) The turbine blade as claimed in claim 11, wherein the platform extends on both the pressure and suction sides of the blade leaf.

18. (previously presented) A gas turbine, comprising:

a flow duct extending along an axis of the turbine having an annular cross section for a working medium; and

a plurality of blade stages having a plurality of annularly arranged turbine blades that extend radially into the flow duct arranged one after another along the axis of the turbine, wherein each turbine blade comprises:

a blade axis perpendicular to the turbine axis,

a blade tip,

a blade root arranged radially opposite the blade tip,

a blade platform arranged adjacent to the blade root and extending transverse to the blade axis,

a blade profile having an airfoil shape arranged between the blade tip and the blade root, wherein:

a first platform wall formed from a resilient elastic metal sheet arranged along a transition from the blade profile to the blade platform that does not structurally support the blade profile and has along a width of the platform an aerodynamically advantageous curved shape, and

a second platform wall arranged along a transition from the blade profile to the blade platform that structurally supports and is continuous with the blade profile and has, along the width of the platform, a set-back step with respect to the first platform and a retaining stop to retain the first platform wall.

19. (previously presented) The gas turbine as claimed in claim 18, wherein during the rotary operation of a rotating turbine blade a centrifugal force acting radially outward from the root of the blade profile toward the blade tip is generated as a result of the blade rotation, the resilient elastic sheet metal first platform wall is pressed against the retaining stop by the centrifugal force and is fastened in place by centrifugal force.

20. (previously presented) The gas turbine as claimed in claim 18, wherein during the operation of a turbine blade in the form of a stationary guide blade, a pressure drop from the blade root toward the blade tip is generated by a cooling medium, the resilient elastic sheet metal first platform wall is pressed against the retaining stop by the pressure drop and thereby fastened by the resulting pressure.

21. (new) A turbine blade, comprising:

a blade leaf arranged along a blade axis having a blade tip, a root opposite the tip, a suction side and a pressure side;

a platform region arranged at the root of the blade leaf;

a platform arranged at the platform region having a width and extends transversely with respect to the blade axis;

a first platform wall arranged along the transition from the blade leaf to the platform that does not structurally support the blade leaf and has along the width of the platform an aerodynamic shape; and

a second platform wall arranged in the platform region that structurally supports and is continuous with the blade leaf and has, along the width of the platform, a set-back step with respect to the first platform;

wherein a portion of the first platform wall is configured to be pressed against a stop of the blade leaf based on a centrifugal force aligned with the blade axis, said portion of the first platform wall aligned at a non-orthogonal angle with respect to the blade axis;

wherein the second platform wall is a continuous extension of the blade leaf;

and wherein the first platform wall is made from an elastic material such that the aerodynamic shape is configured to adapt to a flow of working medium on a hot-gas side of the first platform wall and to a flow of cooling medium on a rear side and within a cooling space between the first and second platform wall, wherein said hot-gas side is opposite to the rear side.